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Peterka, III et al.

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(54) **METHODS AND APPARATUS FOR PERFORMING PROPULSION OPERATIONS USING ELECTRIC PROPULSION SYSTEMS**

(58) **Field of Classification Search**

CPC B64G 1/405; B64G 1/409; B64G 1/24
USPC 244/171.5, 171.1, 171.2, 171.6, 164
See application file for complete search history.

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(51) **Int. Cl.**
B64G 1/40 (2006.01)
B64G 1/00 (2006.01)
B64G 1/64 (2006.01)

(52) **U.S. Cl.**
CPC **B64G 1/405** (2013.01); **B64G 1/002** (2013.01); **B64G 1/40** (2013.01); **B64G 1/409** (2013.01); **B64G 1/641** (2013.01)

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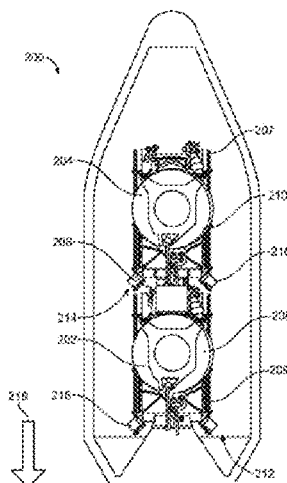
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(57) **ABSTRACT**

Methods and apparatus to methods and apparatus for performing propulsion operations using electric propulsion system are disclosed. An example apparatus includes a frame, a power source coupled to the frame and a payload coupled to the frame, the payload to receive or transmit data. The apparatus also includes an electric propulsion system coupled to the frame. The electric propulsion system is to enable attitude control, momentum control, and orbit control of the apparatus.

33 Claims, 19 Drawing Sheets



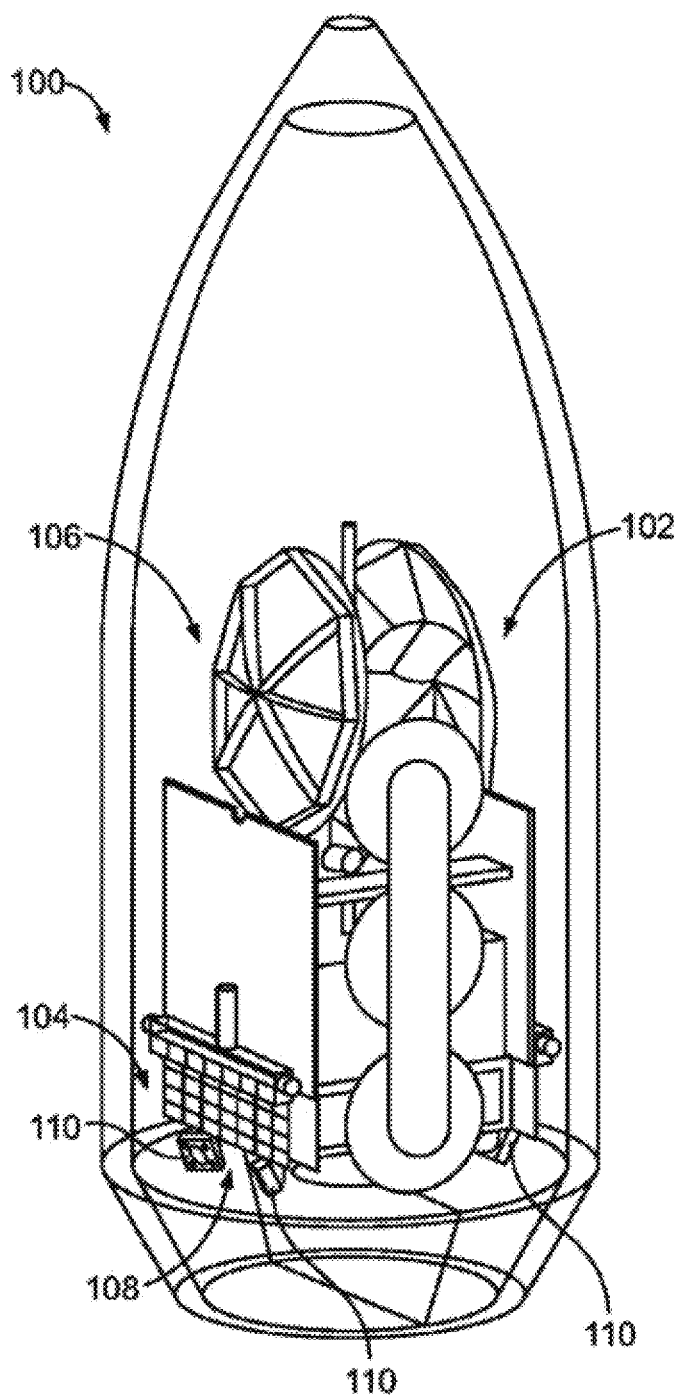


FIG. 1

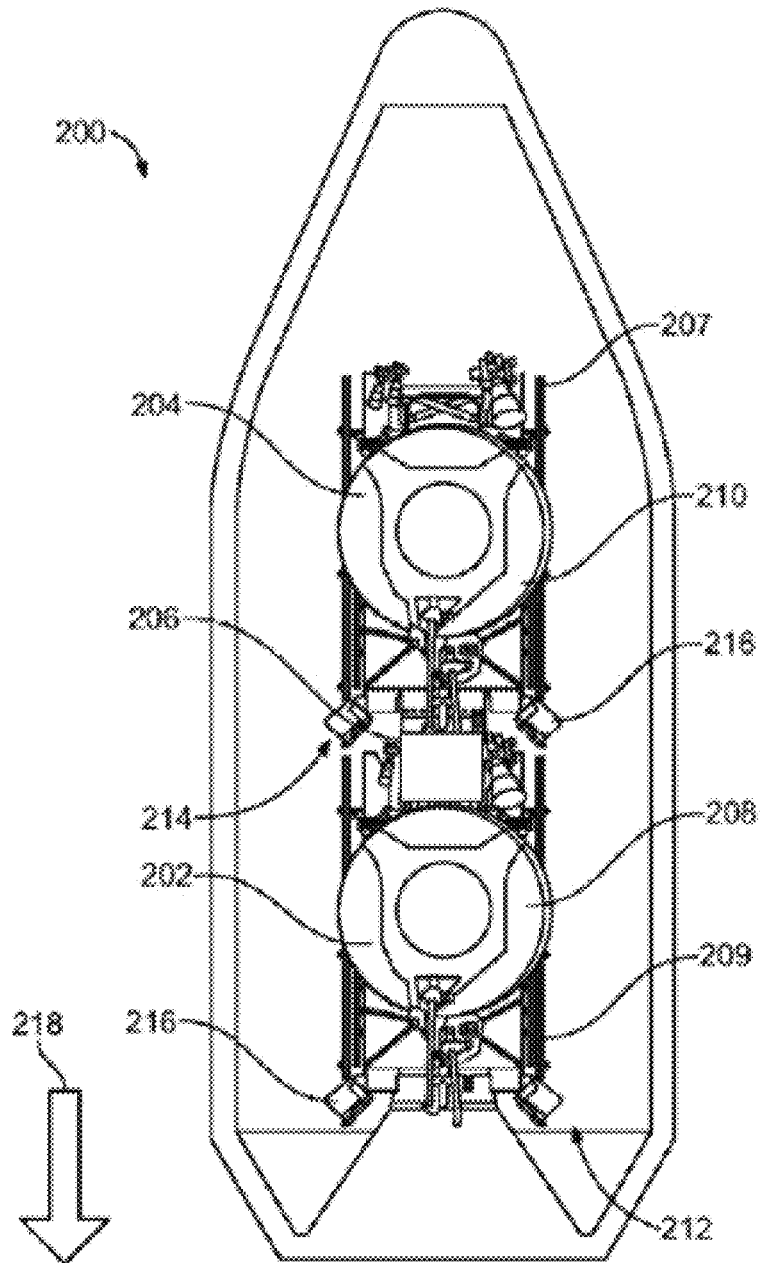


FIG. 2

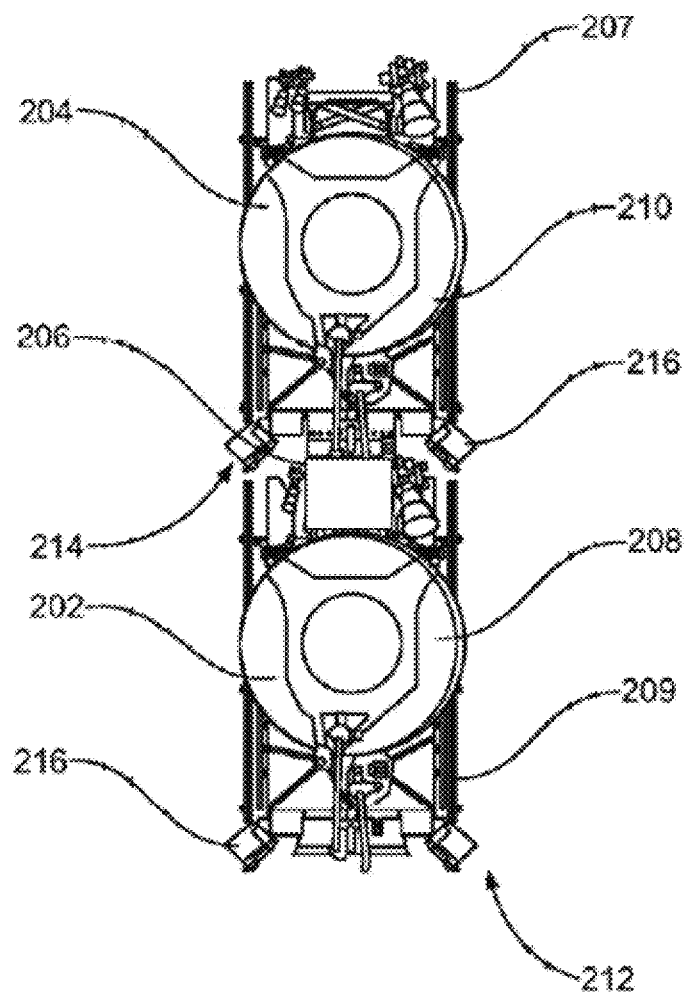


FIG. 3

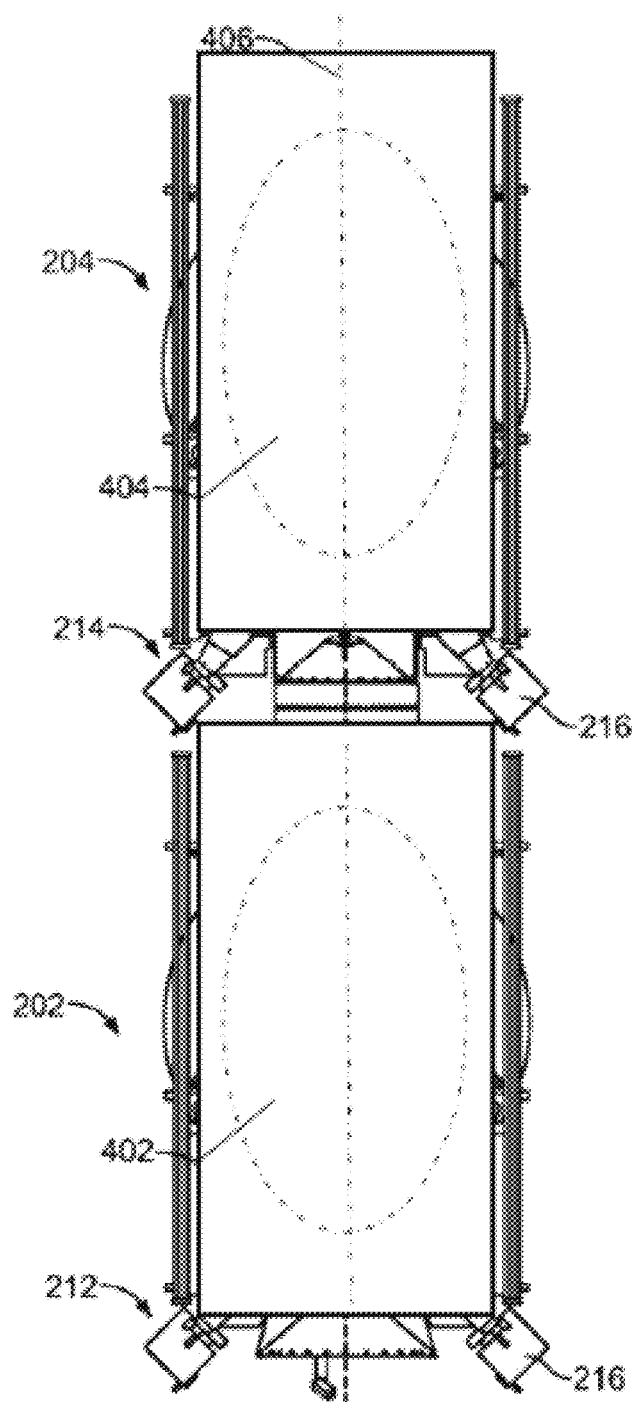


FIG. 4

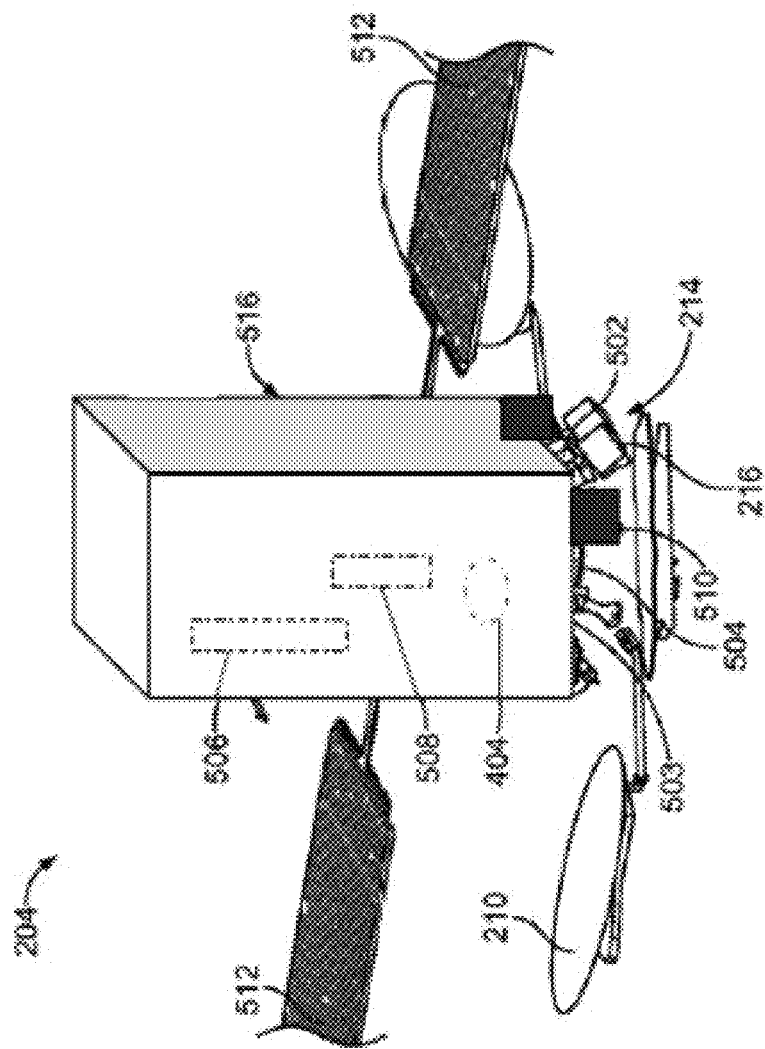


FIG. 5

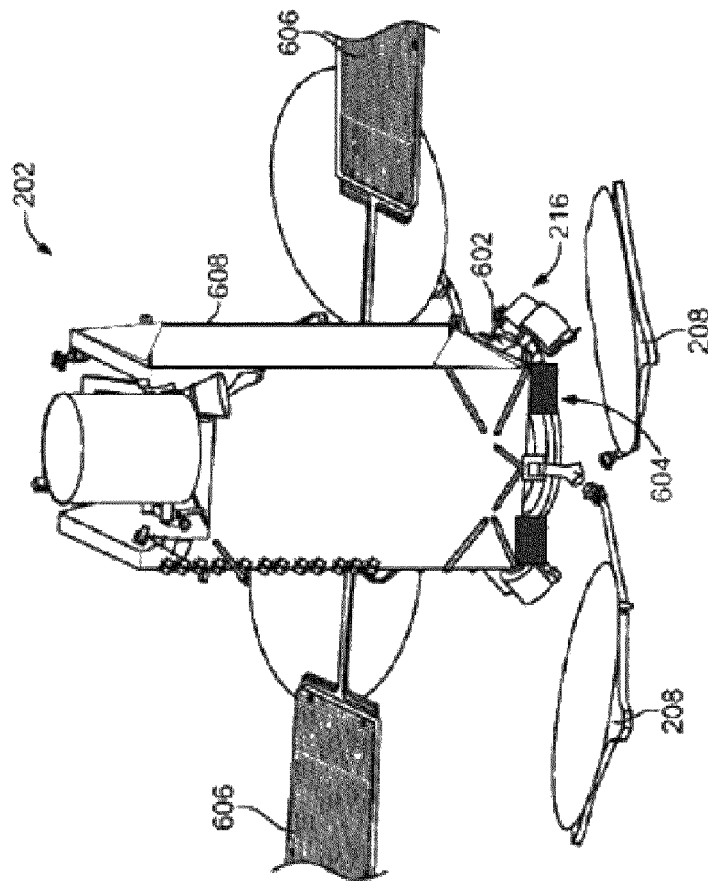


FIG. 6

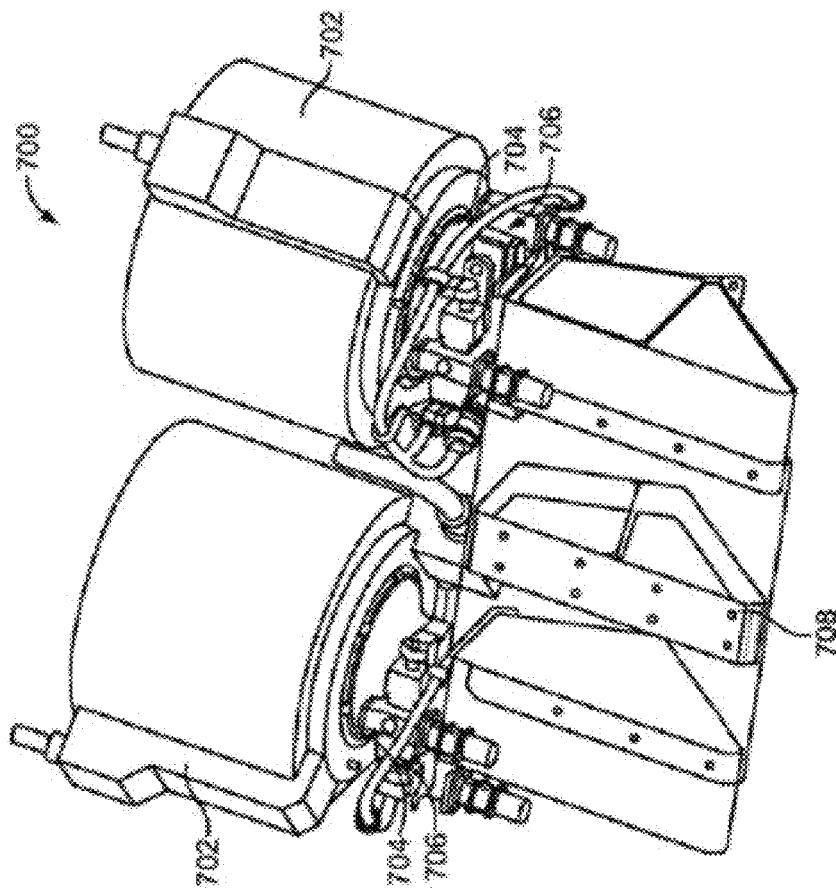


FIG. 7

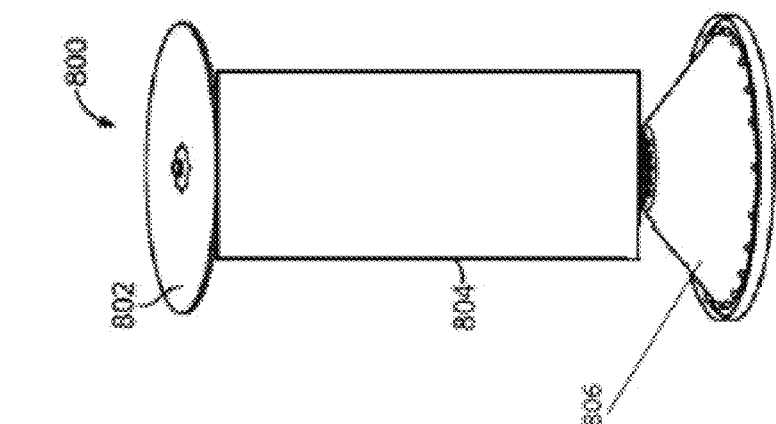


FIG. 8

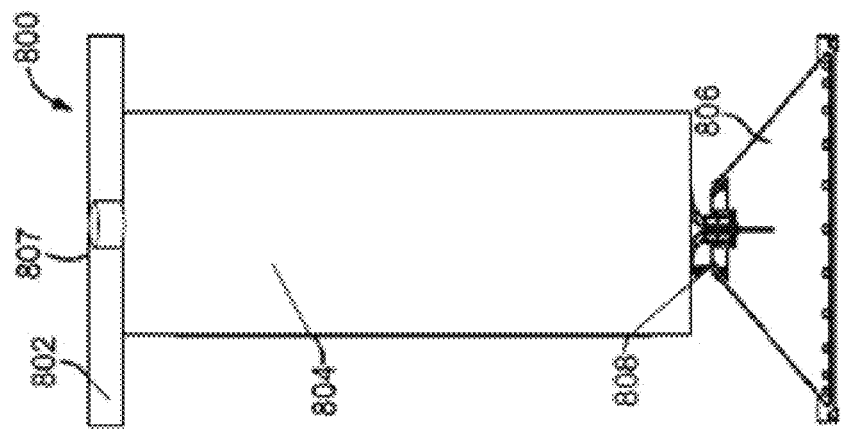


FIG. 9

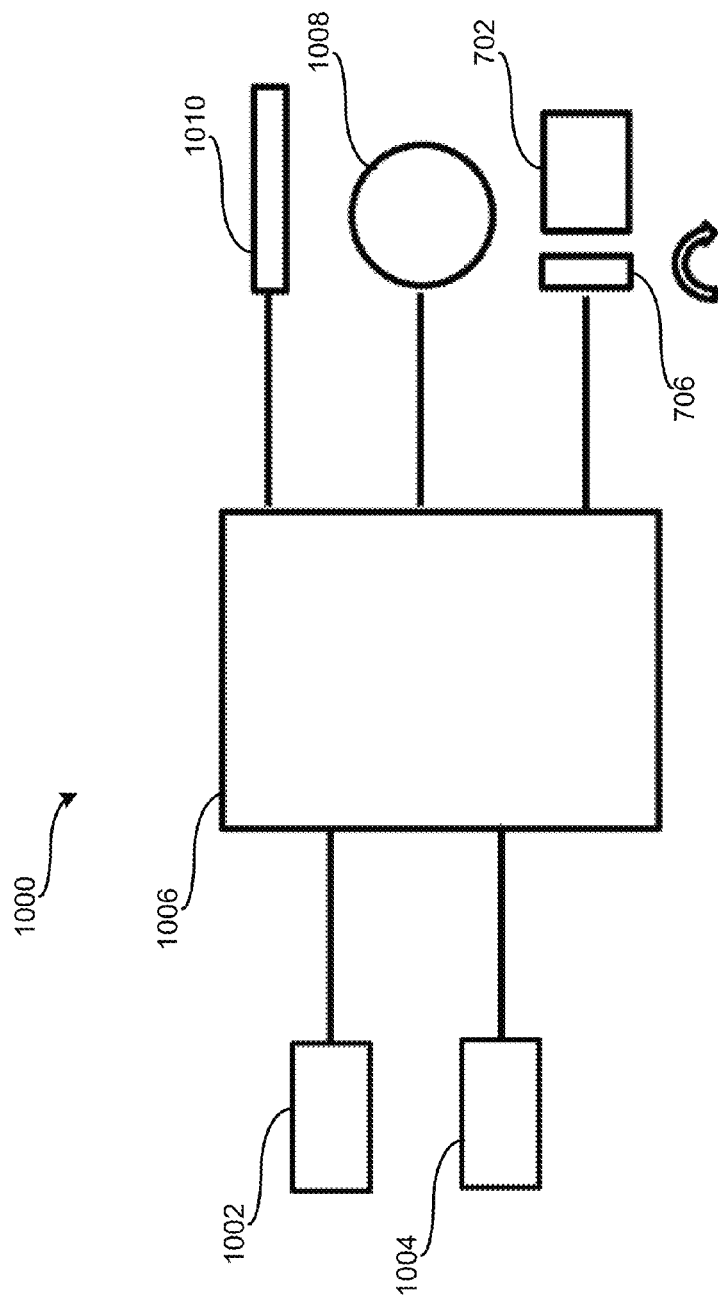


FIG. 10

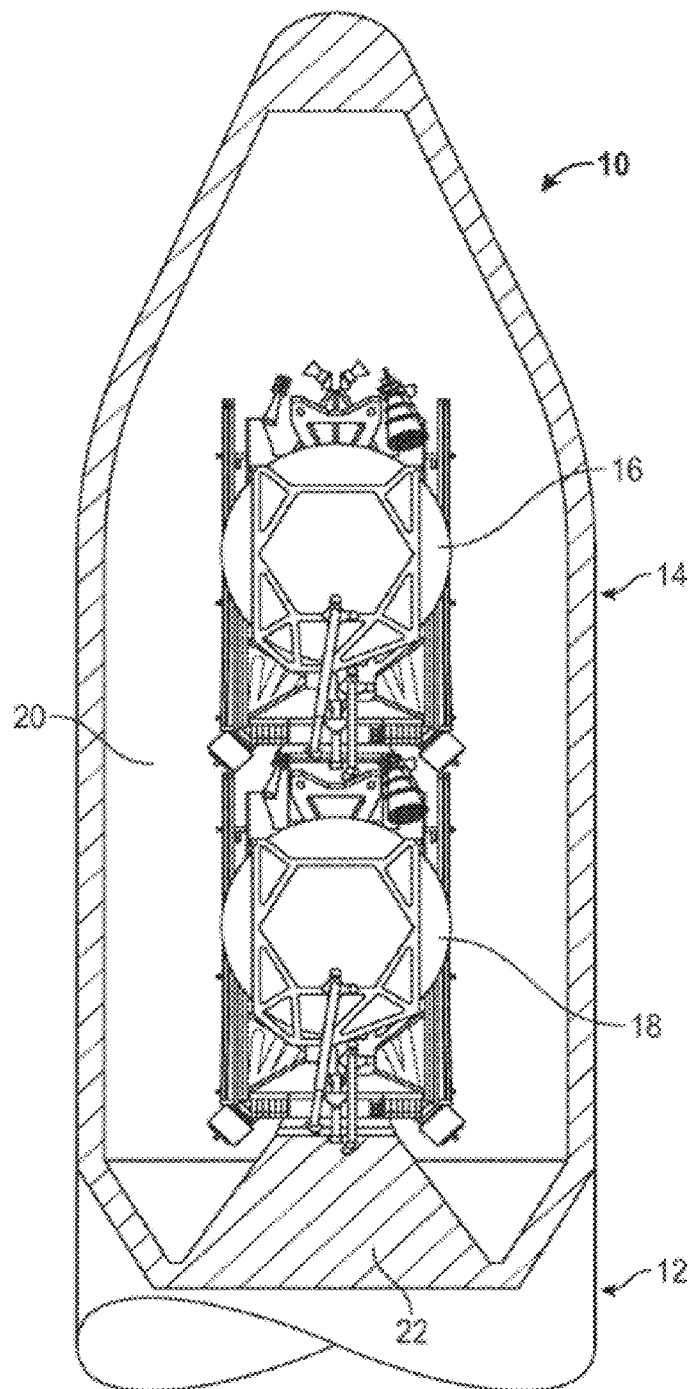


FIG. 11

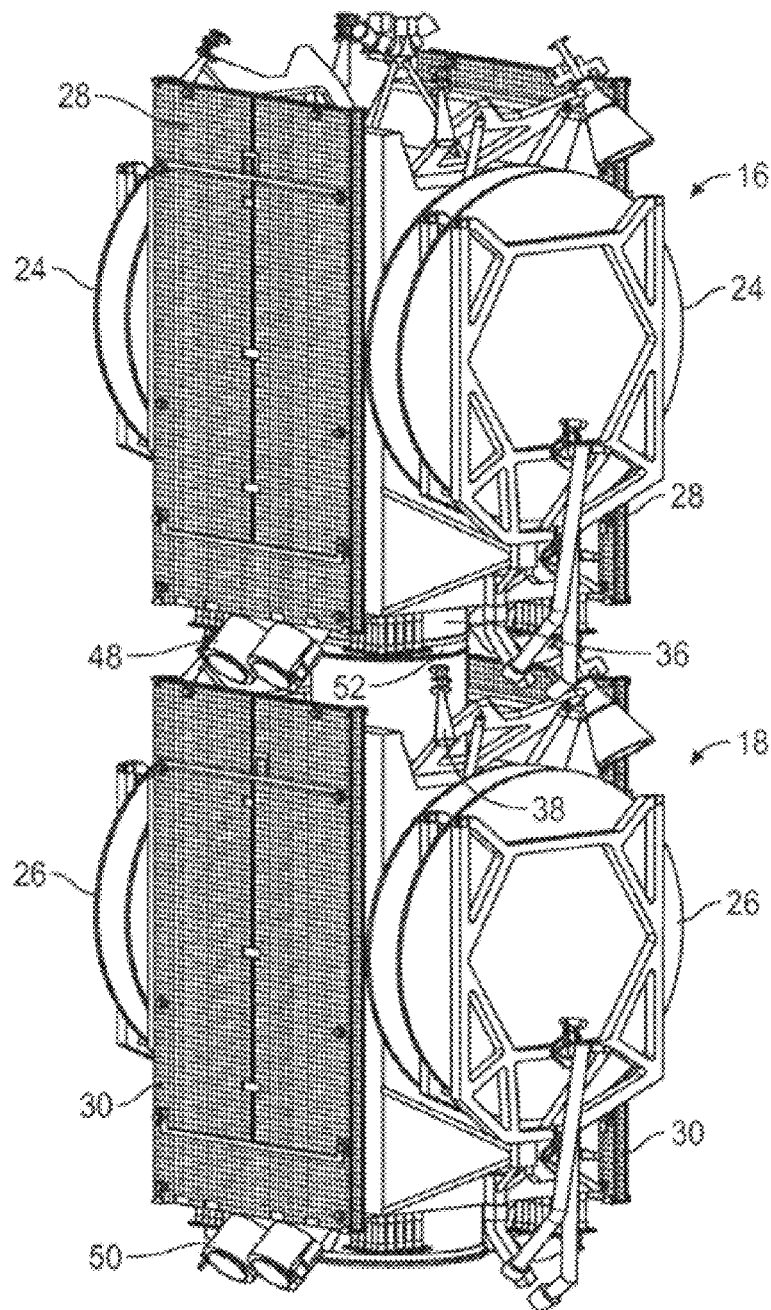


FIG. 12

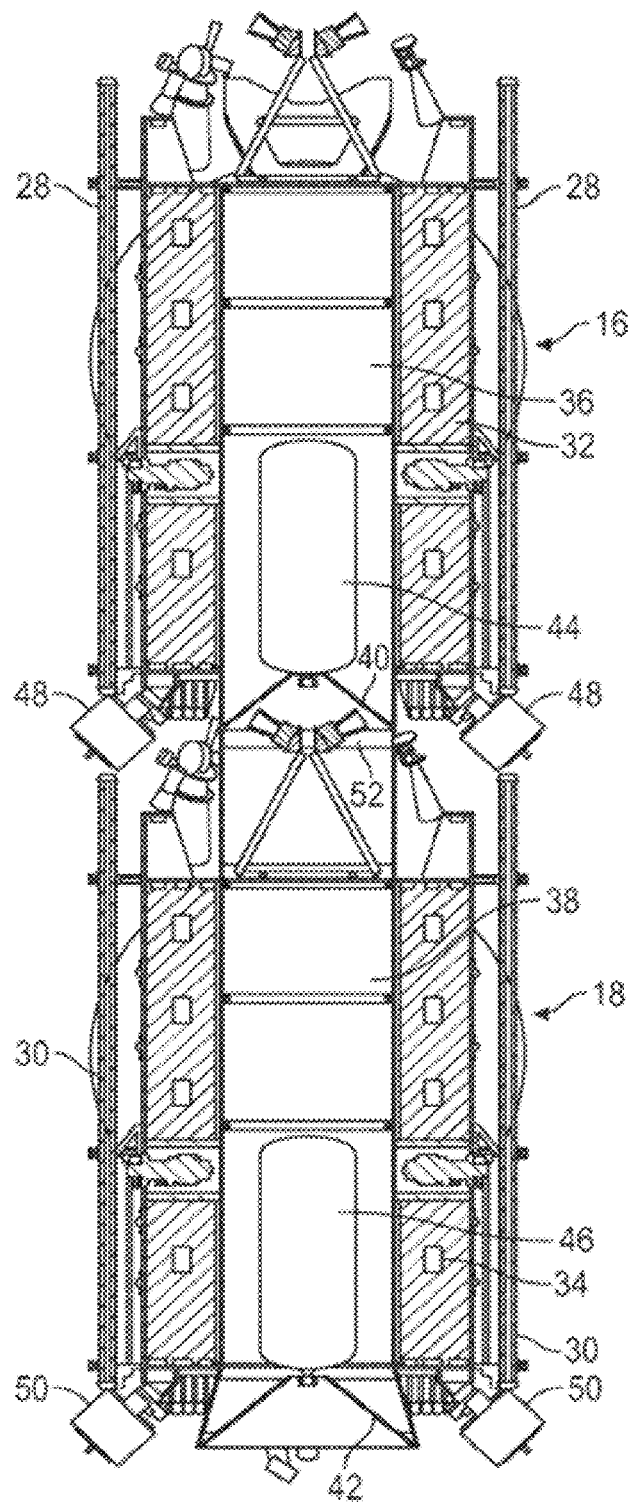


FIG. 13

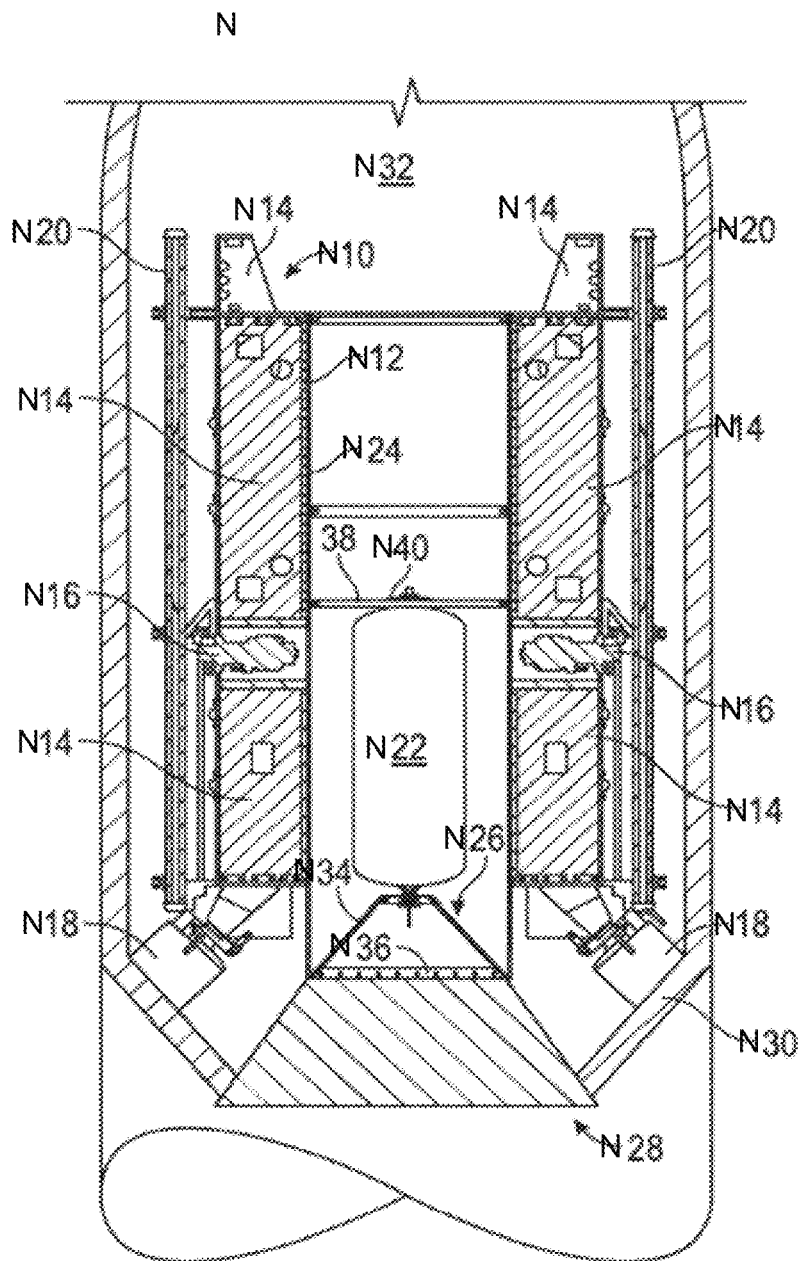


FIG. 14

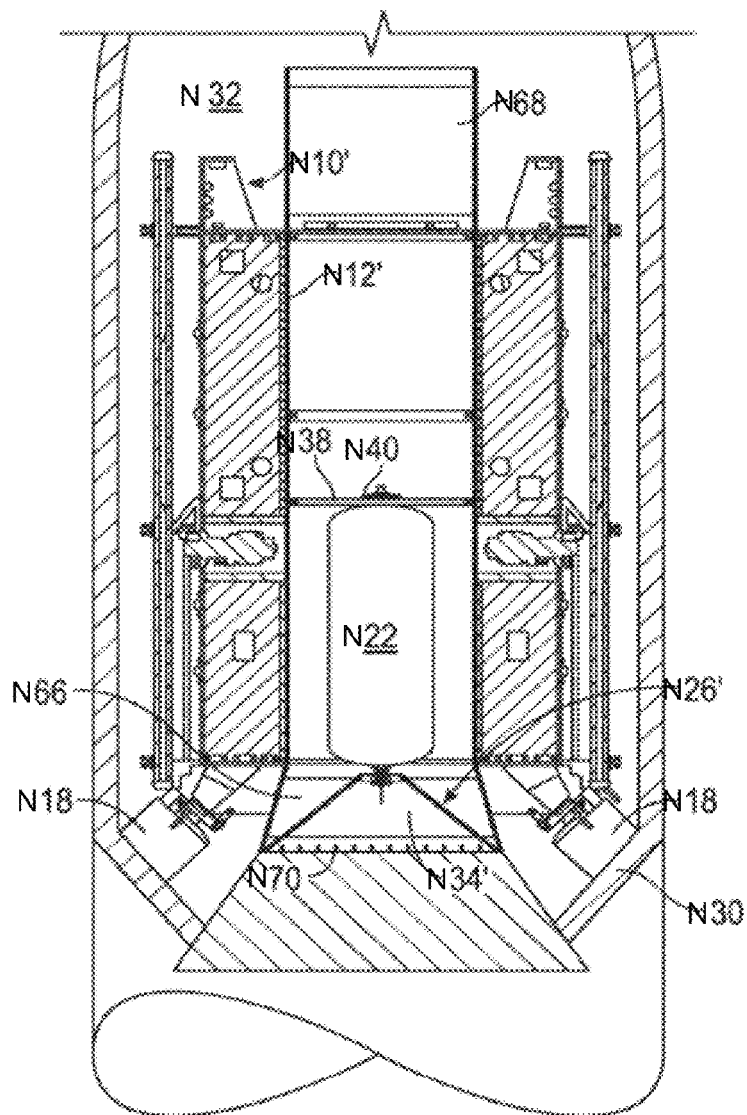


FIG. 15

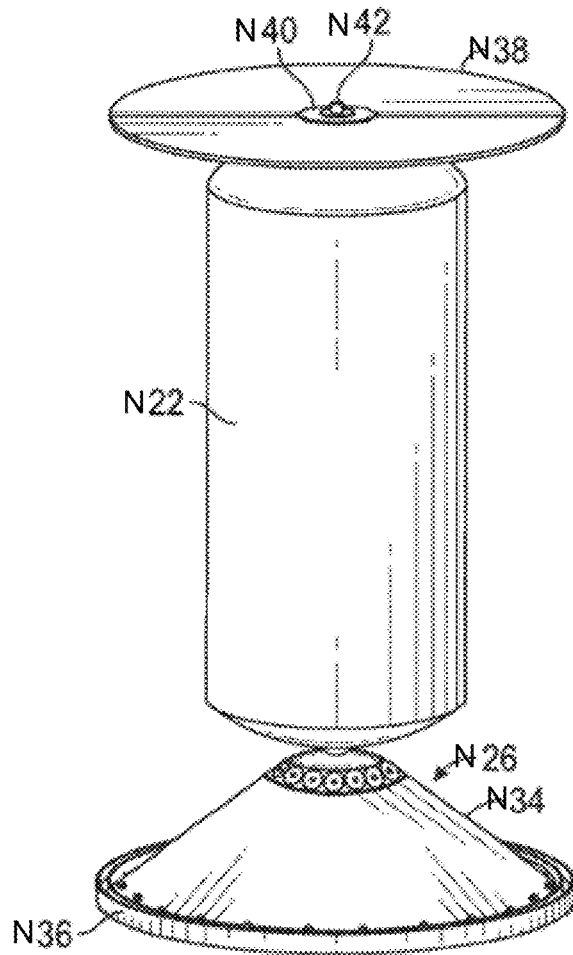


FIG. 16

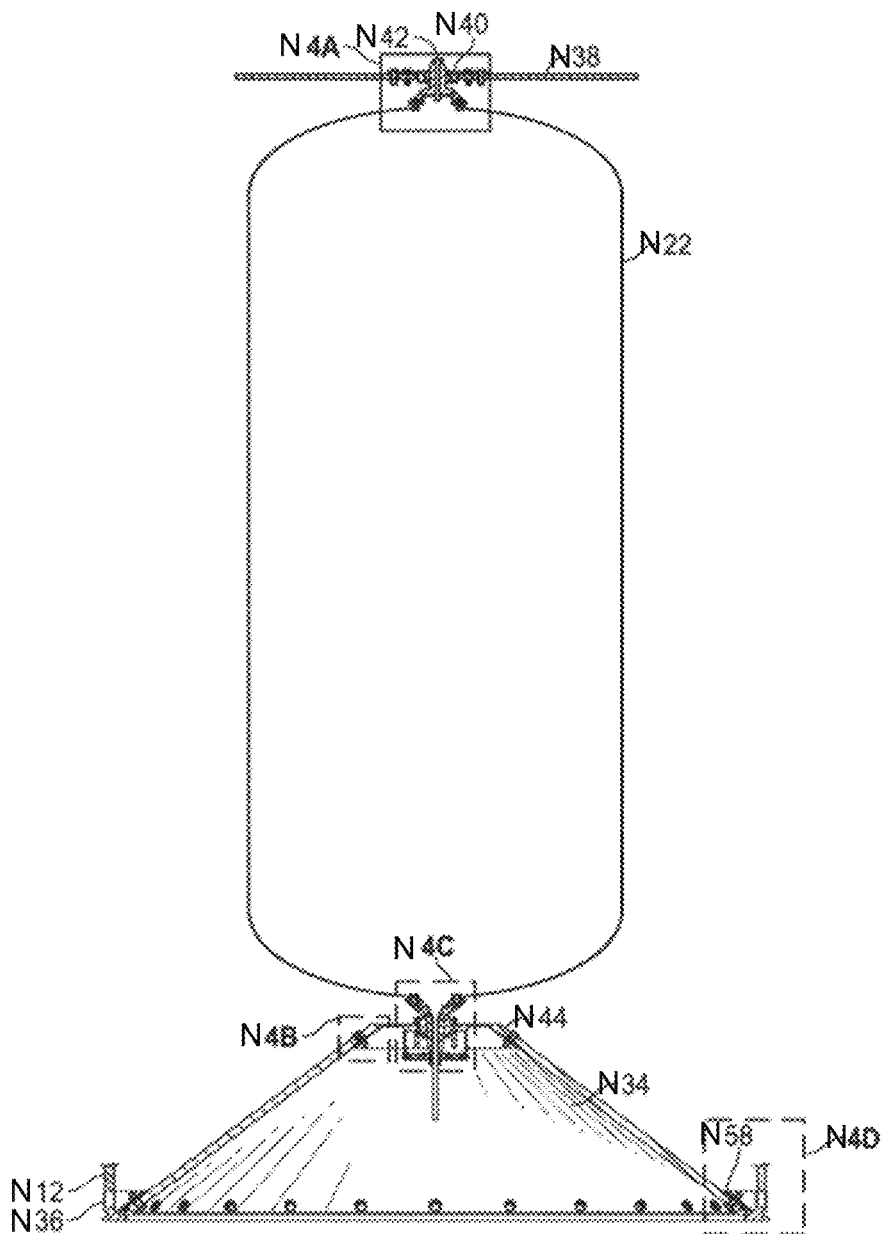


FIG. 17

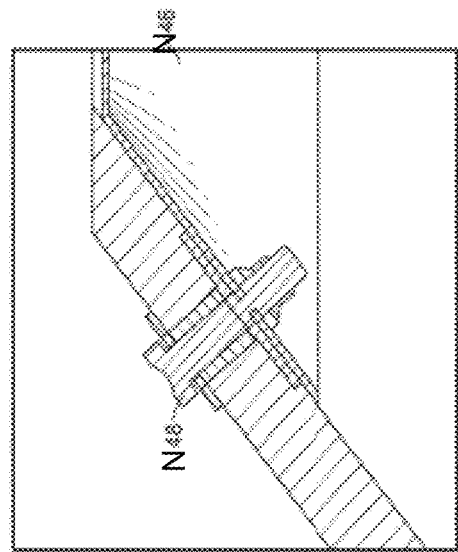


FIG. 19

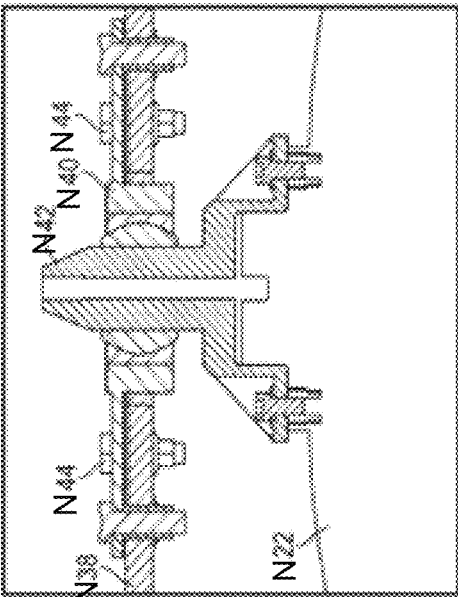


FIG. 18

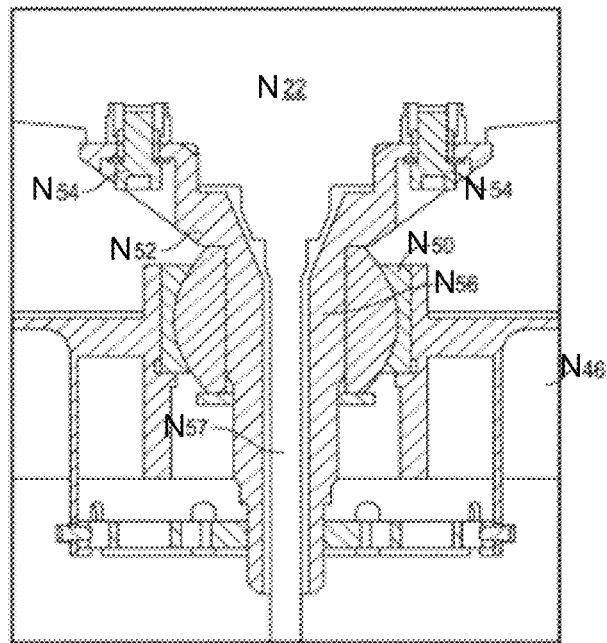


FIG. 20

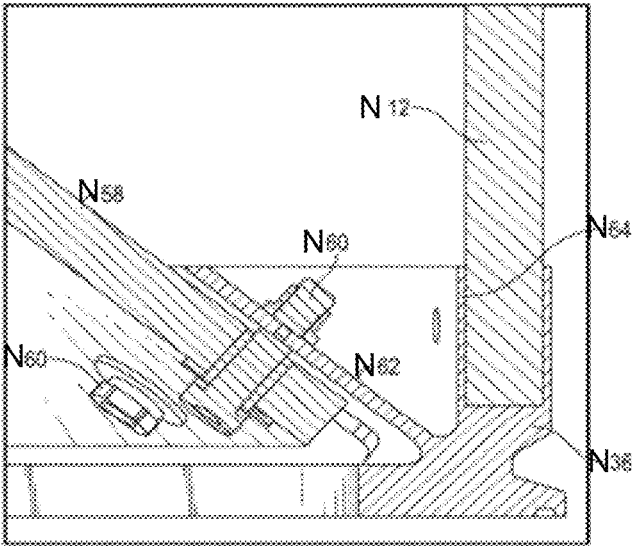


FIG. 21

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METHODS AND APPARATUS FOR PERFORMING PROPULSION OPERATIONS USING ELECTRIC PROPULSION SYSTEMS

RELATED APPLICATIONS

This patent is a continuation in part and claims priority to and incorporates by reference in its entirety U.S. patent application entitled "Multiple Space Vehicle Launch System" filed Sep. 5, 2012, having Ser. No. 13/604,050 with inventors Richard W. Aston, Anna M. Tomzynska, and Glenn Caplin, which claims priority to U.S. Provisional patent application Ser. No. 61/646,222 filed May 11, 2012; and U.S. patent application entitled "Space Propellant Tank Mount" filed on Oct. 15, 2012, having Ser. No. 13/652,101 with inventors Richard W. Aston, Brett Cope, Michael J. Langmack, and Anna M Tomzynska, each of which are hereby incorporated herein by reference in their entireties.

FIELD OF THE DISCLOSURE

This patent relates to electric propulsion systems and, more specifically, to methods and apparatus for performing propulsion operations using electric propulsion systems.

BACKGROUND

Spacecrafts and/or satellites may perform propulsion operations in space. Some of these propulsion operations may include attitude and momentum control, orbit raising, orbit insertion and maintenance, orbit repositioning and/or de-orbit maneuvers. Other propulsion operations may include escaping orbits for interplanetary or extra-solar system missions and/or injection maneuvers to initiate orbit around another planet, moon, etc. Thrust is achieved by acceleration of propellants. Propellants may be accelerated substantially by pressure differences (cold or hot gas systems, (e.g., cold gas)), chemical reactions (including catalytic decomposition, e.g. hydrazine monopropellant systems, hypergolic bipropellant reactions, solid rockets, etc.), and electrical and magnetic interactions (including ion propulsion systems, stationary plasma systems, Hall effect thrusters, magneto-plasma thrusters, etc., (e.g., electric propulsion)). To perform these propulsion operations, some satellites use cold gas systems, or chemical systems, or combinations of cold gas and chemical systems, or combinations of cold gas, chemical and electric propulsion systems.

SUMMARY

An example apparatus in accordance with the teachings of this disclosure includes a frame, a power source coupled to the frame and a payload coupled to the frame. The payload to receive or transmit data. The apparatus also includes an electric propulsion system coupled to the frame. The electric propulsion system is to enable attitude control, momentum control, and orbit control of the apparatus.

Another apparatus includes a launch vehicle and a spacecraft to be positioned in the launch vehicle. The spacecraft includes a frame, a power source coupled to the frame and a payload coupled to the frame. The payload is to receive or transmit data. Another apparatus includes an electric propulsion system coupled to the frame to enable substantially all propulsion operations to be performed without another propulsion system.

Another apparatus includes a launch vehicle, a first module and a second module. The first module is to be removably

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coupled to the second module. The first and second modules are to be positioned in the launch vehicle. The second module includes a frame, a power source coupled to the frame and a payload coupled to the frame. The payload is to receive or transmit data. The apparatus includes an electric propulsion system coupled to the frame. The electric propulsion system is to enable attitude control, momentum control, and orbit control of the second module.

An example method to improve performance of a propulsion system includes using an electric propulsion system coupled to a frame and allowing the electric propulsion system to enable attitude control and orbit control.

The features, functions, and advantages that have been discussed can be achieved independently in various embodiments or may be combined in yet other embodiments further details of which can be seen with reference to the following description and drawings.

BRIEF DESCRIPTION OF THE DRAWINGS

FIG. 1 depicts an example launch vehicle and an example spacecraft in accordance with the teachings of this disclosure.

FIG. 2 depicts an example launch vehicle and example first and second spacecrafts and/or modules in accordance with the teachings of this disclosure.

FIG. 3 depicts an isometric view of the example first and second spacecrafts and/or modules of FIG. 2.

FIG. 4 depicts a cross-sectional view of the example first and second spacecrafts and/or modules of FIG. 2.

FIG. 5 depicts one of the spacecrafts and/or modules of FIG. 2 in a deployed configuration.

FIG. 6 depicts another one of the spacecrafts and/or modules of FIG. 2 in a deployed configuration.

FIG. 7 depicts a portion of an example propulsion unit in accordance with the teachings of this disclosure.

FIGS. 8 and 9 depict an example tank assembly in accordance with the teachings of this disclosure.

FIG. 10 depicts an example attitude control system in accordance with the teachings of this disclosure.

FIG. 11 is a schematic, side elevation in section of an example of the multiple space vehicle launch system of the present disclosure.

FIG. 12 is a schematic, perspective view of two space vehicles depicted in FIG. 11.

FIG. 13 is a schematic, side elevation in section of the space vehicles depicted in FIG. 11.

FIG. 14 is a side elevation in section of an example of a spacecraft incorporating the disclosed tank mount, shown mounted in a launch vehicle fairing.

FIG. 15 is a side elevation in section of a second example of a spacecraft incorporating the disclosed tank mount, shown incorporated in a larger launch vehicle fairing.

FIG. 16 is a perspective view of the disclosed tank support of FIG. 14.

FIG. 17 is a side elevation in section of disclosed tank support structure of FIG. 14.

FIGS. 18, 19, 20, and 21 are enlarged detail views, in section, of portions of the tank support structure shown in FIG. 17.

DETAILED DESCRIPTION

Certain examples are shown in the above-identified figures and described in detail below. In describing these examples, like or identical reference numbers are used to identify the same or similar elements. The figures are not necessarily to scale and certain features and certain views of the figures may

be shown exaggerated in scale or in schematic for clarity and/or conciseness. Additionally, several examples have been described throughout this specification. Any features from any example may be included with, a replacement for, or otherwise combined with other features from other examples.

The examples disclosed herein relate to satellite modules, satellites and/or spacecrafts that use electric propulsion systems (e.g., ion propulsion system, plasma propulsion system such as stationary plasma propulsion systems, Hall Effect propulsion system, etc.) and/or attitude control systems for all propulsion operations and/or mission requirements. The propulsion maneuvers or operations and/or mission requirements may include attitude and momentum control, orbit raising, orbit insertion and maintenance, orbit repositioning, de-orbit maneuvers, etc. Attitude and momentum control includes controlling position and rates of various elements of the spacecraft about each axis, exchange of momentum between spacecraft elements, and utilization and responding to external torques, momentum management, etc. Propulsive maneuvers may provide linear acceleration to change and control orbital parameters such as orbit raising and/or lowering, change in inclination, etc.

By using such an ion propulsion system, the mass of the satellite at launch may be reduced by hundreds of kilograms, the mission life may be extended and/or the processing, production and/or launch costs may be reduced. For example, reducing the mass of the satellite reduces the cost of the launch vehicle. Additionally, using such an electric propulsion system enables the satellite to be repositioned into different orbits and/or orbital slots using substantially less propellant than if a chemical and/or hybrid propulsion system were used.

In contrast, some known satellites use chemical propellants in chemical reaction control for propulsion operations. Other known satellites use electric propulsion in combination with chemical propellants and/or cold gas propulsion systems. However, the cost of the example electric propulsion system is significantly less than some of these known systems (e.g., chemical and electric propulsion systems) due to their complex design and/or the cost of procurement, assembly, integration and/or testing.

By not using chemical propulsion systems, the propulsion systems of the examples disclosed herein eliminate chemical loading operations during the satellite design qualification, acceptance and/or pre-launch preparation at the subsystem level and/or the system level. Thus, electric propulsion systems, which typically use an inert gas propellant, may be installed early in the satellite manufacturing process (e.g., at the fuel tank level) at conventional processing facilities without the risk of exposure to hazardous materials and/or without violating hazardous material processing laws in any state and/or country.

Additionally, by not using chemical propulsion systems, the propulsion systems of the examples disclosed herein are not hazardous to humans and/or the environment and are not limited to being loaded at the launch site in controlled areas. In some examples, chemical propulsion systems use chemicals that are caustic, highly reactive, explosive and/or toxic. Thus, spacecraft fueling operations of chemical propulsion systems are conducted using hazardous materials (HAZMAT) suits and/or other fueling equipment at the launch site. For example, when fueling chemical propulsion systems, special fueling stands, rooms, protection and/or fuel certification may be used and/or needed. Chemical propulsion systems may also expose hardware of the spacecraft to

hazardous materials, such as during fueling operations, which require the hardware to be cleaned and/or refurbished prior to re-use, for example.

FIG. 1 depicts an example single launch vehicle **100** in which a spacecraft, unmanned spacecraft and/or satellite **102** is positioned in its stowed configuration. In this example, the spacecraft **102** includes power sources **104**, payloads **106** and an electric propulsion system **108**. The power sources **104** may include a solar array, radioactive thermionuclear generators (RTGs), energy beaming or transfer apparatus, solar thermal, nuclear thermal, etc. The payloads **106** may include an antenna, a radio frequency (RF) receiver/transmitter, an optical receiver/transmitter, a LASER receiver/transmitter, a light detection and ranging (LIDAR) receiver/transmitter, RADAR receiver/transmitter and/or imaging/detecting systems. The electric propulsion system **108** includes thrusters (e.g., four ion thrusters) **110** that carry out all propulsion operations, including momentum control, orbit raising, orbit insertion and maintenance, orbit repositioning, de-orbit maneuvers, etc. Thus, the spacecraft **102** does not include other propulsion systems, such as chemical or cold gas propulsion systems.

FIG. 2 depicts an example dual launch vehicle **200** in which first and second example spacecrafts, unmanned spacecrafts, modules (e.g., re-entry modules), satellite modules, and/or satellites **202**, **204** are positioned in their stowed configuration. Within the launch vehicle **200**, the first and second spacecrafts **202**, **204** are stacked and/or coupled using an example interface **206**. The interface **206** enables the spacecrafts **202**, **204** to separate after the spacecrafts **202**, **204** are deployed from the launch vehicle **200**, for example. Although FIG. 2 illustrates two stacked spacecraft, it is possible to stack more than two spacecraft and/or alternatively arrange them in other configurations, such as in a side-by-side configuration.

The spacecrafts **202**, **204** include respective frames **207**, **209** that include accommodations for and/or couplings for antennas **208**, **210**. The spacecrafts **202**, **204** also include electric propulsion systems **212**, **214** having thrusters (e.g., four Xenon ion thrusters, ion thrusters) **216** that carry out some or all momentum control, and all orbit raising, orbit insertion and maintenance, orbit repositioning, de-orbit maneuvers, etc. To control the position of the spacecrafts **202**, **204** once deployed from the launch vehicle **200**, the thrusters **216** are independently movable and/or rotatable relative to the respective spacecraft **202**, **204** and/or the frames **207**, **209**. For example, to thrust the first spacecraft **202** forward, the thrusters **216** may be positioned in a direction generally indicated by arrow **218**.

FIGS. 3 and 4 depict the stacked spacecrafts **202**, **204** deployed from the launch vehicle **200**. Alternatively, rather than deploying the stack from the launch vehicle **200**, each spacecraft **202**, **204** may be deployed separately from the launch vehicle **200**, with the upper spacecraft **204** deployed prior to the lower spacecraft **202**. Referring to FIG. 4, the example propulsion systems **212**, **214** include tanks (e.g., Xenon tanks) **402**, **404** that may be positioned along a longitudinal axis **406** of the spacecrafts **202**, **204**. In operation, the respective tanks **402**, **404** are coupled to and provide propellant to one or more of the thrusters **216** to perform the various propulsion operations. The propellant provided may be an inert gas such as Xenon, Argon, Krypton, etc., or other species, such as Mercury. While the example spacecrafts **202**, **204** each have a single tank **402**, **404**, any other number of tanks (e.g., 2, 3, 4, etc.) may be used and may be similarly or differently placed within the respective spacecrafts **202**, **204**.

FIG. 5 depicts the second spacecraft **204** in the deployed configuration. The second spacecraft **204** includes the thrust-

ers **216** that are pivotably mounted to supports **502** and the tank **404** that is coupled to and/or supported by a support and/or tank support **504**. In some examples, the second spacecraft **204** includes an attitude control system and/or reaction and/or momentum control wheels **503** that cooperate to provide attitude control and/or momentum storage for the second spacecraft **204**. In some examples, the attitude control system **503** is an electrically powered attitude control system such as a magnetic torque rod(s), a magnetic torque ring(s) and/or an electrically driven electro-mechanical attitude control system which may be a combination of control wheels and magnetic rods/rings. In some examples, the propulsion system **214** and/or the thrusters **216** include Xenon ion propulsion (XIP) thrusters, power controllers, tankage, flow control and cross-strapping units. In some examples, the second spacecraft **204** also includes an electronics module **506**, payload equipment **508** and a battery and power controller **510**. The second spacecraft **204** may also include the antennas **210**, solar panels and/or arrays **512** and thermal radiators and/or equipment panels **516**.

The electronics module **506** may include spacecraft control electronics, flight software and/or telemetry and command radio frequency (RF) units. The attitude control system **503** may include attitude sensors (e.g., earth sensors, sun sensors, star trackers), Inertial Reference Units (IRUs) or other attitude sensors, reaction wheels or momentum wheels, torque rods or magnetic torquers, etc. The battery and power controller **510** may include battery cells, e.g. Lithium-Ion cells, packs and power controllers. The antennas **210** may receive and/or transmit data and may include active and passive units, antenna structure, a deployment mechanism and/or an antenna positioning mechanism. The solar arrays **512** may include one or more panels, covered in whole or in part with solar cells. The thermal radiators and equipment panels **516** may include a single north oriented radiator and/or equipment, a single south oriented radiator and/or equipment and heat pipes. While the above examples describe the second spacecraft **204** as including particular elements and/or a particular number of those elements, the second spacecraft **204** may include different and/or different quantities of elements.

FIG. 6 depicts the first spacecraft **202** in the deployed configuration. The first spacecraft **202** includes the thrusters **216** that are pivotably mounted to supports **602**. In some examples, the propulsion system **214** and/or the thrusters **216** include Xenon ion propulsion (XIP) thrusters, power controllers, tankage, flow control and cross-strapping units. In some examples, the first spacecraft **202** also includes an electronics module, an attitude control system, and a battery and power controller **604**. The first spacecraft **202** may also include the antennas **208**, solar panels and/or arrays **606** and thermal radiators and equipment panels **608**. While the above examples describe the first spacecraft **202** as including particular elements and/or a particular number of those elements, the first spacecraft **202** may include different and/or different quantities of elements.

FIG. 7 depicts a portion **700** of an example propulsion unit that can be used to implement the examples disclosed herein. The propulsion unit includes Xenon ion propulsion thrusters **702**, adapters **704**, gimbaled platforms **706** and a support structure **708**. In operation, the gimbaled platforms **706** enable the thrusters **702** to be independently rotated and/or positioned to enable the thrusters **706** to move, thrust and/or rotate the respective spacecrafts **202**, **204**, once deployed from the launch vehicle **200**.

FIGS. 8 and 9 depict an example tank assembly **800** that can be used to implement the examples disclosed herein. The assembly **800** includes a tank support panel **802**, a tank **804**

and a conical support structure **806**. In some examples, an axial slip joint and/or monoball **807** couples the tank support panel **802** to the tank **804** to enable the tank to expand and/or contract. In some examples, a moment free monoball mount **808** couples the tank **804** and the support structure **806**.

FIG. 10 depicts an example spacecraft attitude control system **1000** that together with a propulsion system may be used to implement the substantially all electric propulsion system. The attitude control system **1000** includes attitude sensors **1002** (e.g. earth sensors, sun sensors, star trackers), inertial rate sensors **1004**, a controller **1006** and actuators. The actuators may include momentum storage devices **1008** (e.g. momentum wheels, reaction wheels), magnetic torquers **1010**, and the propulsion unit including the thruster **702**. The thruster **702** may be attached to the gimbal **706**. Other examples may include alternate attitude sensing and control apparatus and/or methods.

In some examples, the attitude control system **1000** may be used for attitude and momentum control, etc. In this example, momentum is exchanged between the body (e.g., the spacecraft body) and the wheels (e.g., stored in the wheels) to control the attitude and rate of the body to point the body in the desired direction. The desired direction may be achieved by sensing the current attitude via sensors **1002** and rates via the inertial rate sensors **1004** and the controller **1006** applying torque to the momentum storage devices **1004** to create the desired pointing direction and rates. The desired direction may be such that the linear acceleration created by the thruster **702** will create the orbital changes requested and/or required (e.g. orbit raising). The control system **1000** may also point the body in a desired direction for the payload. In some examples, when orbital operations are requested and/or required concurrently with payload operations, the position and orientation of the thrusters **702** are chosen such that velocity vectors produced by the thrusters when the body is in the desired payload pointing attitude (e.g. earth pointed), can be combined to create the requested and/or required orbital changes. In some examples, the gimbal **706** is used to position the thrust vector nominally through the spacecraft center of mass. Alternatively, several thrusters may be gimbaled to an appropriate position and fired simultaneously such that net torques of all thrusters is zero.

Momentum created by external torques may be periodically 'dumped' from the wheels with the propulsion system by using the gimbal **706** to rotate the thrusters **702** away from the center of mass to achieve the desired torque and/or interaction with solar torques, the earth's magnetic field via magnetic torquers **1010**, etc. In some examples, the example thruster **702** and the gimbal **706** arrangement uses the same set of thrusters in different orientations depending on the requested and/or required direction of acceleration and payload pointing. Other examples may include additional thrusters with little or no gimbal capability dedicated to creating acceleration in primarily a single direction. Other control systems may use the thruster **702** to create the desired spacecraft rate and pointing ('thrust vector steered') rather than rate and pointing through exchange of momentum. Other examples of the attitude control system **1002** may be an electrical, mechanical and/or magnetic system and/or any other suitable system.

As disclosed herein, the example electric propulsion system may use and/or be associated with ionized gas expulsion, non-ionized gas expulsion and/or cold gas states. For example, the example electric propulsion system may be used to propel the apparatus in either an ionized state and/or a non-ionized state of the species. In examples in which the electric propulsion system uses non-ionized gas, the gas may

be discharged and/or dribbled out of the thrusters and/or the apparatus may include one or more cold gas thrusters. The cold gas thrusters may be used for occasional control and/or contingency operations of the apparatus.

In some examples, the non-ionized gas and/or the cold gas thrusters may be used in association with de-orbit operations and/or other phases of a mission. For example, the cold gas thrusters may use propellant from, for example, the tank 402 and/or the 404 and/or another gas tank (e.g., xenon tank) and feed system and/or another cold gas type.

As set forth herein, an example apparatus includes a frame a power source coupled to the frame and a payload coupled to the frame. The payload is to receive or transmit data. The apparatus includes an electric propulsion system coupled to the frame. The electric propulsion system is to enable attitude control, momentum control, and orbit control of the apparatus. In some examples, orbit control includes orbit maintaining, orbit changing, orbit raising, orbit insertion, orbit re-positioning, and de-orbit maneuvers of the apparatus. In some examples, momentum control includes momentum management. In some examples, the apparatus also includes a controller to control the electric propulsion system. In some examples, the apparatus also includes the electric propulsion system includes a thruster. In some examples, the electric propulsion system includes a gimbaled platform to enable the thruster to move relative to the frame. In some examples, the electric propulsion system includes a plurality of thrusters. In some examples, each of the thrusters is independently movable. In some examples, the electric propulsion system includes a tank to receive propellant. In some examples, the tank is positioned along a longitudinal axis of the frame. In some examples, the power source includes a solar array fixed or movable between a stowed configuration and a deployed configuration. In some examples, the payload is at least partly movable between a stowed configuration and a deployed configuration. In some examples, in a stowed configuration, the apparatus is to be positioned in a launch vehicle. In some examples, the electric propulsion system includes a Xenon ion propulsion system, a plasma propulsion system such as a stationary plasma thruster, or a Hall Effect propulsion system.

Another example apparatus includes a launch vehicle and a spacecraft to be positioned in the launch vehicle. The spacecraft includes a frame, a power source coupled to the frame and a payload coupled to the frame. The payload to receive or transmit data. An electric propulsion system is coupled to the frame to enable substantially all propulsion operations to be performed without another propulsion system. In some examples, the electric propulsion system includes a thruster and a gimbaled platform. The gimbaled platform is to enable the thruster to move relative to the frame. In some examples, the electric propulsion system includes a plurality of thrusters. In some examples the electric propulsion system includes a tank to receive propellant. In some examples, the tank is positioned along a longitudinal axis of the frame. In some examples, substantially all propulsion operations includes attitude control, momentum control, and orbit control of the apparatus. In some examples, orbit control includes orbit maintaining, orbit changing, orbit raising, orbit insertion, orbit re-positioning, and de-orbit maneuvers of the spacecraft.

Another apparatus includes a launch vehicle, a first module and a second module. The first module is to be removably coupled to the second module. The first and second modules are to be positioned in the launch vehicle. The second module includes a frame, a power source coupled to the frame and a payload coupled to the frame. The payload is to receive or transmit data. The apparatus includes an electric propulsion

system coupled to the frame. The electric propulsion system is to enable attitude control, momentum control, and orbit control of the second module. In some examples, the electric propulsion system includes a tank to receive propellant. In some examples, the tank is positioned along a longitudinal axis of the frame.

An example method to improve performance of a propulsion system includes using an electric propulsion system coupled to a frame and allowing the electric propulsion system to enable attitude control and orbit control. In some examples, using the electric propulsion system includes using a plurality of independently movable thrusters. In some examples, using the electric propulsion system includes using a propellant stored in a tank positioned along a longitudinal axis of the frame. In some examples, the method also includes allowing the electric propulsion system or an attitude control system to enable momentum control. In some examples, using the electric propulsion system comprises using a Xenon ion propulsion system, a plasma propulsion system such as a stationary plasma thruster, or a Hall Effect propulsion system.

The examples disclosed herein relate to a multiple space vehicle launch system that may include a first space vehicle, a second space vehicle releasably attached to the first space vehicle and oriented relative to the first space vehicle such that, when placed within a fairing, a launch load from the first space vehicle is transmitted to and borne by the second space vehicle, thereby eliminating the need for Sylde or other reinforcing or support structure. In an example, the first and second space vehicles each may include one of an electrical propulsion motor and a hybrid chemical and electrical propulsion motor. By utilizing electrical propulsion motors in the space vehicles, the total mass of the space vehicle may be significantly reduced when compared to a space vehicle having a chemical propulsion motor, which may enable support structures such as Sylde to be eliminated.

According to an example, a multiple space vehicle launch system may include a first space vehicle, a second space vehicle releasably attached to the first space vehicle and oriented relative to the first space vehicle such that when placed within a fairing, a launch load from the first space vehicle is transmitted to and borne by the second space vehicle. The first and second space vehicles each may include one of an electrical propulsion unit and a hybrid chemical and electrical propulsion unit.

In another example, a spacecraft launch system may include a launch vehicle with a fairing having a payload region, and a plurality of space vehicles disposed within the payload region. The plurality of space vehicles may be oriented in a vertically stacked manner such that at least a portion of gravitational and launch loads of an upper space vehicle are transmitted to and borne by a lower space vehicle. Each of the space vehicles may include at least one of an electrical propulsion unit and a hybrid electrical and chemical propulsion unit.

In yet another example, a method of launching a plurality of space vehicles may include providing a plurality of space vehicles, each of the plurality of space vehicles including at least one of an electrical propulsion unit and a hybrid electrical and chemical propulsion unit, orienting the plurality of space vehicles in a stacked manner within a payload region of a fairing of a launch vehicle such that gravitational and launch loads of an upper one of the plurality of space vehicles is transmitted to and borne by a lower one of the plurality of space vehicles, and launching the launch vehicle with the plurality of space vehicles.

In the examples described above and others, the use of traditional inter-launch vehicle fairing, Sylde, and inter-fair-

ing separation systems may be eliminated. This reduces the non-revenue generating payload mass and may reserve more available mass for revenue generating payload.

As shown in FIG. 11, an example multiple space vehicle launch system, generally designated 10, is used with a launch vehicle 12 having a fairing 14. The system 10 may include a first or upper space vehicle, generally designated 16, and a second or lower space vehicle, generally designated 18. The space vehicles are positioned within a payload region 20 of the fairing 14. It should be noted that, although FIG. 11 shows a space vehicle launch system 10 having two space vehicles 16, 18, it is within the scope of the disclosure to provide a space vehicle launch system having three or more space vehicles.

Regardless of the number of space vehicles 16, 18 employed in the launch system, the arrangement of space vehicles within the fairing 14 may be in a stacked, vertical configuration as shown in FIG. 11. The term "vertical" as used herein refers to the orientation of the stacked space vehicles 16, 18 relative to a launch pad supporting the launch vehicle 12 when the launch vehicle is oriented in a vertical position, or a vertically stacked manner, relative to the Earth. In an example, the stacked space vehicles 16, 18 may be aligned with, and may coincide with, a central longitudinal axis of the fairing 14 and/or launch vehicle 12. The lower space vehicle 18 may rest upon a base 22 that may be a part of the fairing 14.

As shown in FIGS. 12 and 13, the space vehicles 16, 18 may be satellites. In various examples, the space vehicles 16, 18 may be geosynchronous satellites, interplanetary probes, combinations thereof, or any type of space vehicle having a propulsion system that is launched by a launch vehicle 12 (FIG. 11).

The space vehicles 16, 18 may include antenna reflectors 24, 26, respectively, and deployable solar arrays 28, 30, respectively. As best shown in FIG. 13, the space vehicles 16, 18 may include shear load panels 32, 34 that are mounted on core structure 36, 38, respectively.

The core structures 36, 38 may be cylindrical in shape and hollow. Core structures may be of other shapes and not depart from the scope of this disclosure. Core structure 36 may be made of a strong, light material such as graphite, and in one example have a wall thickness of 0.09". Core structure 38 also may be made of a strong, light material such as graphite, and in one example have a wall thickness of 0.45". The shear panels 32, 34 may support the solar arrays 28, 30 of the space vehicles 16, 18, respectively.

In the example shown in FIGS. 12 and 13, the space vehicles 16, 18 each may include an electric propulsion motor, generally designated 40, 42, respectively. Electric propulsion motors 40, 42 may include an ion/plasma motor that utilizes Xenon gas as a propellant that is stored in tanks 44, 46 that may be positioned within core structure 36, 38, respectively. The electric propulsion motors 40, 42 also may include exhaust nozzles 48, 50, respectively.

In the examples shown in FIGS. 12 and 13, the space vehicles 16, 18 each may include a single electric propulsion motor 40, 42 that may constitute the sole source of propulsion and navigation for that space vehicle; no other propulsion source may be included. The components 40, 42 of space vehicles 16, 18 also may represent other types of electric propulsion motors, as well as hybrid electric/chemical propulsion motors. It is also within the scope of the disclosure to provide space vehicle 16 with an electric propulsion motor 40 and provide space vehicle 18 with a hybrid electric/chemical propulsion motor 42. Use of electric propulsion motors 40, 42, or hybrid electric/chemical propulsion motors may be

advantageous because they reduce the overall mass of the space vehicles 16, 18 in comparison to chemical propulsion motors.

In one example, the upper space vehicle 16 may be connected to the lower space vehicle 18 by a pre-tensioned release band 52 that connects the core structure 36 of the upper vehicle with the core structure 38 of the lower vehicle. As shown in the figures, the core structure 38 of the lower vehicle 18 may extend upwardly above the upper edge of the solar arrays 30 of the lower vehicle to engage the core structure 36 which, in the example shown, may not extend beyond the lower edge of the solar arrays 28 of the upper space vehicle.

In operation, the upper and lower space vehicles 16, 18, respectively, first may be attached to each other by the pre-tensioned release band 52. The combined space vehicles 16, 18 may be placed within the fairing 14 of a launch vehicle 12, as shown in FIG. 14, so that the lower space vehicle rests upon the base 22 of the fairing.

When the launch vehicle is standing on the launch pad, the launch vehicle 12, fairing 14 and space vehicles 16, 18 may be oriented vertically relative to the Earth. In this configuration, the downward gravitational force of the upper space vehicle 16 may be transmitted to and borne entirely by the lower space vehicle 18. In the example shown, this gravitational force may be transmitted entirely from the core structure 36 of the upper space vehicle 16 to the core structure 38 of the lower space vehicle 18.

During liftoff of the launch vehicle 12, the acceleration forces of the upper space vehicle 16 likewise may be transmitted through the core structure 36 to the core structure 38 of the lower space vehicle 18. In the example shown, the upper and lower space vehicles 16, 18 may be linearly and vertically aligned in a vertically stacked configuration so that the gravitational and launch loads of the upper space vehicle 16 are efficiently transmitted to and borne entirely by the lower space vehicle 18.

In conclusion, two configuration features of the disclosed space vehicle launch system combine to provide a reduction in overall launch system mass. First, the individual space vehicles do not use conventional chemical propellant, but instead use electric propulsion, in one example, which has a higher efficiency and thus requires significantly less propellant mass. In another example, the space vehicles may use a hybrid electric/chemical propulsion motor. Second, the space vehicles may be stacked, one on top of the other, so that the launch loads from the upper space vehicle may pass through the lower space vehicle.

The upper and lower space vehicles may include a compatible mounting structure for releasably mounting adjacent spacecraft. This structure may eliminate the need for an inner fairing structure or a fairing separation system, which otherwise might be necessary for multiply manifested spacecraft. The disclosed vehicle launch system may eliminate a significant amount of mass that is not required to fulfill the primary spacecraft mission, which allows more available mass for revenue-generating payload. Further, minimizing propellant mass and non-functional structure mass from the launch vehicle optimizes the overall system mass.

As set forth herein, an example multiple space vehicle launch system includes a first space vehicle, a second space vehicle releasably attached to the first space vehicle and oriented relative to the first space vehicle such that when placed within a fairing, a launch load from the first space vehicle is transmitted to and borne by the second space vehicle. The first and the second space vehicles each include one of an electrical propulsion unit and a hybrid chemical and electrical prop-

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pulsion unit. In some examples, the first and the second space vehicles are oriented in a stacked configuration. In some examples, the first and the second space vehicles are oriented in a vertically stacked configuration during launch. In some examples, the example multiple space vehicle launch system also includes a fairing shaped to enclose the first and the second space vehicles.

In some examples, the fairing includes a base shaped to support the second space vehicle. In some examples, the second space vehicle is attached to the first space vehicle such that the launch load from the first space vehicle is transmitted entirely to, and borne entirely by, the second space vehicle. In some examples, the first space vehicle includes a first core structure. The second space vehicle includes a second core structure; and the first core structure is attached to the second core structure. In some examples, the launch load from the first space vehicle is transmitted to the second space vehicle through the first core structure and the second core structure. In some examples, the first and the second space vehicles each include an electrical propulsion unit. In some examples, the electrical propulsion unit is an ion/plasma propulsion unit. In some examples, the electrical propulsion unit includes Xenon gas. In some examples, at least one of the first and the second space vehicles is a satellite.

Another example spacecraft launch system includes a launch vehicle including a fairing having a payload region, a plurality of space vehicles disposed within the payload region. The plurality of space vehicles being oriented in a vertically stacked manner such that at least a portion of gravitational and launch loads of an upper space vehicle are transmitted to and borne by a lower space vehicle. Each of the space vehicles includes at least one of an electrical propulsion unit and a hybrid electrical and chemical propulsion unit. In some examples, at least one of the space vehicles is a satellite. In some examples, each of the space vehicles includes an electrical propulsion unit.

An example method of launching a plurality of space vehicles includes providing a plurality of space vehicles, each of the plurality of space vehicles including at least one of an electrical propulsion unit and a hybrid electrical and chemical propulsion unit, orienting the plurality of space vehicles in a stacked manner within a payload region of a fairing of a launch vehicle such that gravitational and launch loads of an upper one of the plurality of space vehicles is transmitted to and borne by a lower one of the plurality of space vehicles, and launching the launch vehicle with the plurality of space vehicles.

In some examples, providing a plurality of space vehicles includes providing at least one satellite. In some examples, providing a plurality of space vehicles includes providing a first space vehicle and providing a second space vehicle. In some examples, orienting the plurality of space vehicles includes attaching the first space vehicle to the second space vehicle such that a launch load of the first space vehicle is transmitted to and borne by the second space vehicle. In some examples, attaching the first space vehicle to the second space vehicle includes attaching a core structure of the first space vehicle to a core structure of the second space vehicle.

An example multiple space vehicle launch system that may be adapted to be disposed within a payload region of a launch vehicle fairing is disclosed. The launch system may include a first space vehicle, a second space vehicle releasably attached to the first space vehicle and oriented relative to the first space vehicle such that, when placed within the fairing, a launch load of the first space vehicle is transmitted to and borne by the second space vehicle. In certain examples, the first and second space vehicles each may include one of an electrical

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propulsion unit and a hybrid chemical and electrical propulsion unit. Use of electrical or hybrid chemical and electrical propulsion units enables the second space vehicle to bear all or a significant portion of the launch load of the first space vehicle, thereby eliminating the need for additional support structure.

The examples disclosed herein relate to a spacecraft having a primary structural frame and a propellant tank, the spacecraft including a tank mount adapted to engage a portion of the propellant tank, the tank mount being configured to transfer launch loads directly from the propellant tank to the launch vehicle interface ring. In one aspect, the propellant tank mount includes a conical shell having a first end adapted to engage an end of a propellant tank, and a second end adapted to engage a launch vehicle interface ring. In another aspect, a method of mounting a propellant tank to a spacecraft having a primary structural frame includes providing a tank support adapted to engage a portion of the propellant tank, the tank support being configured to transfer launch loads directly from a propellant tank to a launch vehicle interface ring.

In some examples of the disclosed spacecraft, propellant tank mount and method is that the propellant tank may be supported independently of the central thrust tube of the spacecraft. Consequently, the spacecraft and mount may accommodate propellant tanks of a variety of shapes and diameters. The shape and diameter of the propellant tank need not be dictated by the inside diameter of the central thrust tube. In some examples of the disclosed spacecraft, propellant tank mount and method is that the launch load of the propellant tank may be transferred directly from the propellant tank to the launch vehicle interface ring, and not borne by the central thrust tube of the spacecraft.

The disclosed design may provide a mass-efficient solution because the propellant tank load (i.e., the force exerted by the mass of the propellant tank during launch, and when the launch vehicle is accelerating, as a result of acceleration of the launch vehicle and spacecraft) may bypass the spacecraft's primary structure. This may enable use of a simplified and relatively lighter primary structure, so that a larger portion of the available mass of the spacecraft may be allotted to instrumentation and other spacecraft payload.

As shown in FIG. 14, the disclosed spacecraft, generally designated N10, may include a primary structural frame that may be in the form of a cylindrical central thrust tube N12 that extends substantially the entire length of the spacecraft. The thrust tube N12 also may support stiffener panels N14, solar wing drives N16 and thrusters N18. Thrusters N18 may be used for attitude control and/or moving the spacecraft N10 to a different orbit. The solar wing drives N16 may support solar panels N20.

The thrusters N18 may be include electric propulsion units connected to a propellant tank N22. In examples, the thrusters N18 may be gridded electrostatic ion thrusters, or Hall effect thrusters. The propellant tank N22 may contain xenon gas propellant under pressure and may be dimensioned to be spaced from the inner surface N24 of the central thrust tube N12. Although shown in FIG. 14 as having a cylindrical shape, in examples the propellant tank N22 may be spherical, elliptical or oval in shape, etc. In an example, the propellant tank N22 may be a metallic pressure vessel with a composite overwrap reinforcement. In examples, the propellant tank N22 may be made of aluminum or titanium, and may or may not have overwrap reinforcement.

As shown in FIGS. 14 and 16, the spacecraft N10 may include a tank mount, generally designated N26. The tank mount N26 may serve to attach the spacecraft N10 to the base

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N28 of a launch vehicle N30, so that the spacecraft N10 may be positioned within the payload region N32 of the launch vehicle. The tank mount N26 may include a conical shell N34 that may be made of a lightweight, strong composite material. In an example, the material may include graphite or carbon fiber and may have a honeycomb structure. In other examples, the conical shell N34 may be made of metal, such as titanium, steel or aluminum alloy. The conical shell N34 may be attached to a launch vehicle interface ring N36, which may be part of the base N28 of a launch vehicle N30.

At an opposite end of the propellant tank N22, the tank mount N26 may include a forward tank support panel N38. The forward tank support panel N38 may be disk-shaped and sized to engage the inner periphery N24 of the central thrust tube N12. The forward tank support panel N38 may be a solid disk, as shown, or may have voids to reduce weight. The forward tank support panel N38 may be attached to the propellant tank N22 by a pivotal mount N40, such as the monoball bearing axial slip joint shown. Other types of pivotal mounts may be employed. The forward tank support panel N38 may be attached to the inner periphery N24 of the central thrust tube N12 by welding, brazing, adhesives or other means.

As shown in FIGS. 17 and 18, the propellant tank N22 may include an axially extending forward tank boss N42 that may extend through and is captured by the monoball bearing joint N40. The monoball bearing joint N40 may be attached to the forward tank support panel N38 by fasteners such as bolts N44. In other examples, the monoball bearing joint N40 may be attached to the forward tank support panel N38 by a suitable adhesive, by welding, by rivets, or a combination of the foregoing. The monoball bearing joint N40 may be made of metal, such as an aluminum alloy or titanium.

As shown in FIGS. 17 and 19, the upper end N44 of the conical shell N34 may be attached to a cap N46 that may be made of a hardened material such as titanium or other metal. The attaching mechanism may be by adhesives, or bolts N48 as shown in FIG. 19. As shown in FIGS. 17 and 20, the cap N46 may include a pivotal mount N50, such as the moment-free monoball bearing mount shown. Monoball bearing mount N50 may receive an aft tank boss N52 of the propellant tank N22. The aft tank boss N52 may be attached to the propellant tank by screws N54 and may include an adapter tube N56 that extends through and is captured by the monoball bearing mount N50. The tube N56 may be hollow and shaped to receive an outlet tube N57 of the propellant tank N22. In one example, the tube N56 may be slidable relative to the monoball bearing mount to allow for expansion and contraction of the propellant tank N22, and accommodate any out-of-tolerance conditions. Similarly, the forward tank boss 42 (FIG. 18) may be slidably retained by the monoball bearing slip joint N40. In examples, both joints N40 and N50 may allow axial (i.e., in the direction of the longitudinal axis of the spacecraft 10) and pivotal movement of the propellant tank N22 relative to the spacecraft N10, central thrust tube N12 and conical support N34.

As shown in FIGS. 17 and 21, the lower end N58 of the conical shell N34 may be attached to the launch vehicle interface ring N36 by bolts N60 that extend through the lower end and through tabs N62 formed on the interface ring N36. As shown in FIG. 21, the interface ring N36 also may include an angular slot N64 shaped to receive the bottom of the central thrust tube N12 (FIG. 14), and the joint may be secured by means such as an adhesive, welding or brazing, mechanical fasteners such as screws, or a combination of the foregoing.

As shown in FIG. 15, a spacecraft N10' may include a central thrust tube N12' that is flared outwardly at the bottom

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of N66 to accommodate a greater thrust load, in the event that the central thrust tube N12' may support the thrust tube N68 of a second spacecraft to be launched in tandem with the spacecraft N10. In this example, the propellant tank N22 may be supported in a similar fashion as that described with reference to FIG. 14, except that the conical shell N34' may be shaped to flare outwardly at a greater angle than shell N34 engage a larger interface ring N70.

The disclosed spacecraft N10, N10' and tank mount N26, N26' provide a low-cost mounting system that may transfer launch loads from the lower propellant tank nozzle N52 through the conical shell N34 and to the launch vehicle interface ring N36. Therefore, the launch load of the propellant tank N22 may be conveyed directly to the interface ring 36 without transferring a load to the central thrust tube N12. Because the connection between the propellant tank N22 and the forward tank support panel 38 is by way of a slip joint N40, there is not thrust load transmitted to the central thrust tube N12 at that location. Thus, the entire thrust load of the propellant tank may be borne by the interface ring N36 and not the structural frame of the spacecraft N12, N12'. Further, because the propellant tank is attached to the spacecraft N12, N12' at its upper and lower ends by boss N42 and nozzle N56, the support system will accommodate a variety of propellant tank dimensions and diameters.

As set forth herein, an example spacecraft having a primary structural frame and a propellant tank includes a tank mount adapted to engage a portion of the propellant tank. The tank mount being configured to transfer launch loads directly from the propellant tank to a launch vehicle interface ring. In some examples, the tank mount is conical in shape. In some examples, the tank mount includes a conical shell. In some examples, the tank mount includes a pivotal mount attached to the propellant tank and the conical shell. In some examples, the pivotal mount includes a monoball bearing mount.

In some examples, the conical shell is formed of a composite material. In some examples, the conical shell if formed of one or more of graphite, carbon fiber, titanium, steel and aluminum alloy. In some examples, the conical shell has a honeycomb structure. In some examples, the conical shell includes a lower peripheral edge shaped to engage the launch vehicle interface ring. In some examples, the lower peripheral edge is mechanically attached to the launch vehicle interface ring. In some examples, the tank mount includes a plurality of bolts mechanically attaching the lower peripheral edge to the launch vehicle interface ring.

In some examples, the propellant tank is generally one of spherical, elliptical, cylindrical and oval in shape. In some examples, the propellant tank is configured to retain xenon gas propellant. In some examples, the primary structural frame includes a cylindrical central thrust tube, and the propellant tank is shaped to fit within and not contact the cylindrical central thrust tube directly. In some examples, example spacecraft includes a forward tank support panel for supporting an end of the propellant tank opposite the tank support, the forward tank support panel being shaped to engage the primary structural frame. In some examples, the forward tank support panel includes a pivotal mount attached to the propellant tank. In some examples, the pivotal mount includes a monoball bearing. In some examples, the propellant tank includes axially extending forward tank boss, and the monoball bearing is shaped to receive the forward tank boss for relative slidable and pivotal movement.

An example propellant tank mount for a spacecraft includes a conical shell having a first end adapted to engage an end of a propellant tank, and a second end adapted to engage a launch vehicle interface ring.

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An example method of mounting a propellant tank to a spacecraft having a primary structural frame includes providing a tank support adapted to engage a portion of the propellant tank, the tank support being configured to transfer launch loads directly from the propellant tank to a launch vehicle interface ring.

A spacecraft having a primary structural frame and a propellant tank, in which the spacecraft may include a tank mount adopted to engage a portion of the propellant tank, the tank mount being configured to transfer launch loads directly from the propellant tank to a launch vehicle interface ring.

Furthermore, although certain example methods, apparatus and articles of manufacture have been described herein, the scope of coverage of this patent is not limited thereto. On the contrary, this patent covers all methods, apparatus and articles of manufacture fairly falling within the scope of the appended claims either literally or under the doctrine of equivalents.

What is claimed is:

1. An apparatus, comprising:
 - a frame;
 - a power source coupled to the frame;
 - a payload coupled to the frame, the payload to receive or transmit data; and
 - an electric propulsion system coupled to the frame, the electric propulsion system comprising a first thruster coupled to a first side of the frame and a second thruster coupled to a second side of the frame, the first and second thrusters to enable attitude control, momentum control, and orbit control of the apparatus without use of another attitude system or another momentum system, no other propulsion system provided to enable the attitude control, the momentum control, and the orbit control.
2. The apparatus of claim 1, wherein the orbit control comprises orbit maintaining, orbit changing, orbit raising, orbit insertion, orbit re-positioning, and de-orbit maneuvers of the apparatus and wherein the momentum control comprises momentum management.
3. The apparatus of claim 1, further comprising a controller to control the electric propulsion system.
4. The apparatus of claim 1, wherein the electric propulsion system comprises a gimbaled platform to enable the first thruster to move relative to the frame.
5. The apparatus of claim 1, wherein the power source comprises a solar array fixed or movable between a stowed configuration and a deployed configuration.
6. The apparatus of claim 1, wherein the payload is at least partly movable between a stowed configuration and a deployed configuration.
7. The apparatus of claim 1, further comprising a launch vehicle into which the apparatus is to be positioned in a stowed configuration.
8. The apparatus of claim 1, wherein the electric propulsion system comprises a Xenon ion propulsion system, a plasma propulsion system, or a Hall Effect propulsion system.
9. The apparatus of claim 1, wherein the first and second thrusters each comprise a plurality of thrusters.
10. The apparatus of claim 9, wherein each of the thrusters is independently movable.
11. The apparatus of claim 1, wherein the electric propulsion system comprises a tank to receive a propellant, the propellant comprising an inert gas propellant, a Xenon gas propellant, an Argon gas propellant, or a Krypton gas propellant.
12. The apparatus of claim 11, wherein the tank is positioned along a longitudinal axis of the frame.

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13. The apparatus of claim 11, wherein the tank further comprises a cap having a slip joint to enable the tank to expand or contract.

14. The apparatus of claim 11, further comprising a support interface to couple the tank to the frame.

15. The apparatus of claim 14, wherein the support interface is to transfer a launch load from the tank to the frame.

16. The apparatus of claim 14, wherein the support interface comprises a conical support interface.

17. The apparatus of claim 14, wherein the support interface is coupled to the tank via a pivotal mount to reduce a thrust load imparted onto a portion of the frame.

18. An apparatus, comprising:

- a launch vehicle; and
- a spacecraft to be positioned in the launch vehicle, the spacecraft, comprising:
 - a frame;
 - a power source coupled to the frame;
 - a payload coupled to the frame, the payload to receive or transmit data; and
 - an electric propulsion system coupled to the frame, the electric propulsion system comprising a first thruster adjacent a first side of the frame and a second thruster adjacent a second side of the frame, the first thruster and the second thruster to enable propulsion operations to be performed, and wherein no other propulsion system is provided to enable the propulsion operations.

19. The apparatus of claim 18, wherein the electric propulsion system comprises a gimbaled platform, the gimbaled platform to enable the first thruster and the second thruster to move relative to the frame.

20. The apparatus of claim 18, wherein the electric propulsion system comprises a tank to receive a propellant, the propellant comprising an inert gas propellant, a Xenon gas propellant, an Argon gas propellant, or a Krypton gas propellant.

21. The apparatus of claim 20, wherein the tank is positioned along a longitudinal axis of the frame.

22. The apparatus of claim 20, wherein the propulsion operations includes attitude control, momentum control, and orbit control of the apparatus.

23. The apparatus of claim 22, wherein orbit control comprises orbit maintaining, orbit changing, orbit raising, orbit insertion, orbit re-positioning, and de-orbit maneuvers of the spacecraft.

24. An apparatus, comprising:

- a launch vehicle;
- a first module; and
- a second module, the first module to be removably coupled to the second module, the first and second modules to be positioned in the launch vehicle, the second module, comprising:
 - a frame;
 - a power source coupled to the frame;
 - a payload coupled to the frame, the payload to receive or transmit data; and
 - an electric propulsion system coupled to the frame, the electric propulsion system comprising first electric thrusters adjacent a first side of the frame and second electric thrusters adjacent a second side of the frame, the first and second electric thrusters to enable attitude control, momentum control, and orbit control of the second module, and wherein no other propulsion system is provided to enable the attitude control, the momentum control, or the orbit control.

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25. The apparatus of claim 24, wherein the electric propulsion system comprises a tank to receive propellant, the propellant comprising an inert gas propellant, a Xenon gas propellant, an Argon gas propellant, or a Krypton gas propellant.

26. The apparatus of claim 25, wherein the tank is positioned along a longitudinal axis of the frame.

27. The apparatus of claim 24, wherein the first module and the second module are removably coupled to enable the first and second modules to be launched in tandem.

28. A method to improve performance of a propulsion system, comprising:

using an electric propulsion system coupled to a frame, the electric propulsion system comprising first thrusters adjacent a first side of the frame and second thrusters adjacent a second side of the frame; and

allowing the first and second thrusters to enable attitude control and orbit control without using another propulsion system to enable the attitude control and the orbit control, no other propulsion system provided to enable the attitude control and the orbit control.

29. The method of claim 28, wherein using the first and second thrusters comprises using a plurality of independently movable thrusters.

30. The method of claim 28, wherein using the electric propulsion system comprises using a propellant stored in a tank positioned along a longitudinal axis of the frame, the

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propellant comprising an inert gas propellant, a Xenon gas propellant, an Argon gas propellant, or a Krypton gas propellant.

31. The method of claim 28, further comprising allowing the electric propulsion system or an attitude control system to enable momentum control.

32. The method of claim 28, wherein using the electric propulsion system comprises using a Xenon ion propulsion system, a plasma propulsion system, or a Hall Effect propulsion system.

33. An apparatus, comprising:

a frame;

a power source coupled to the frame;

a payload coupled to the frame, the payload to receive or transmit data; and

an electric propulsion system coupled to the frame, the electric propulsion system comprising a first thruster and a second thruster, the first thruster positioned adjacent a first side of the frame and the second thruster positioned adjacent a second side of the frame, the second side spaced from the first side, the first and second thrusters to enable attitude control and momentum control to be performed without use of another attitude system, orbit control system, or momentum system, no other propulsion control system provided to enable the attitude control and the momentum control.

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